

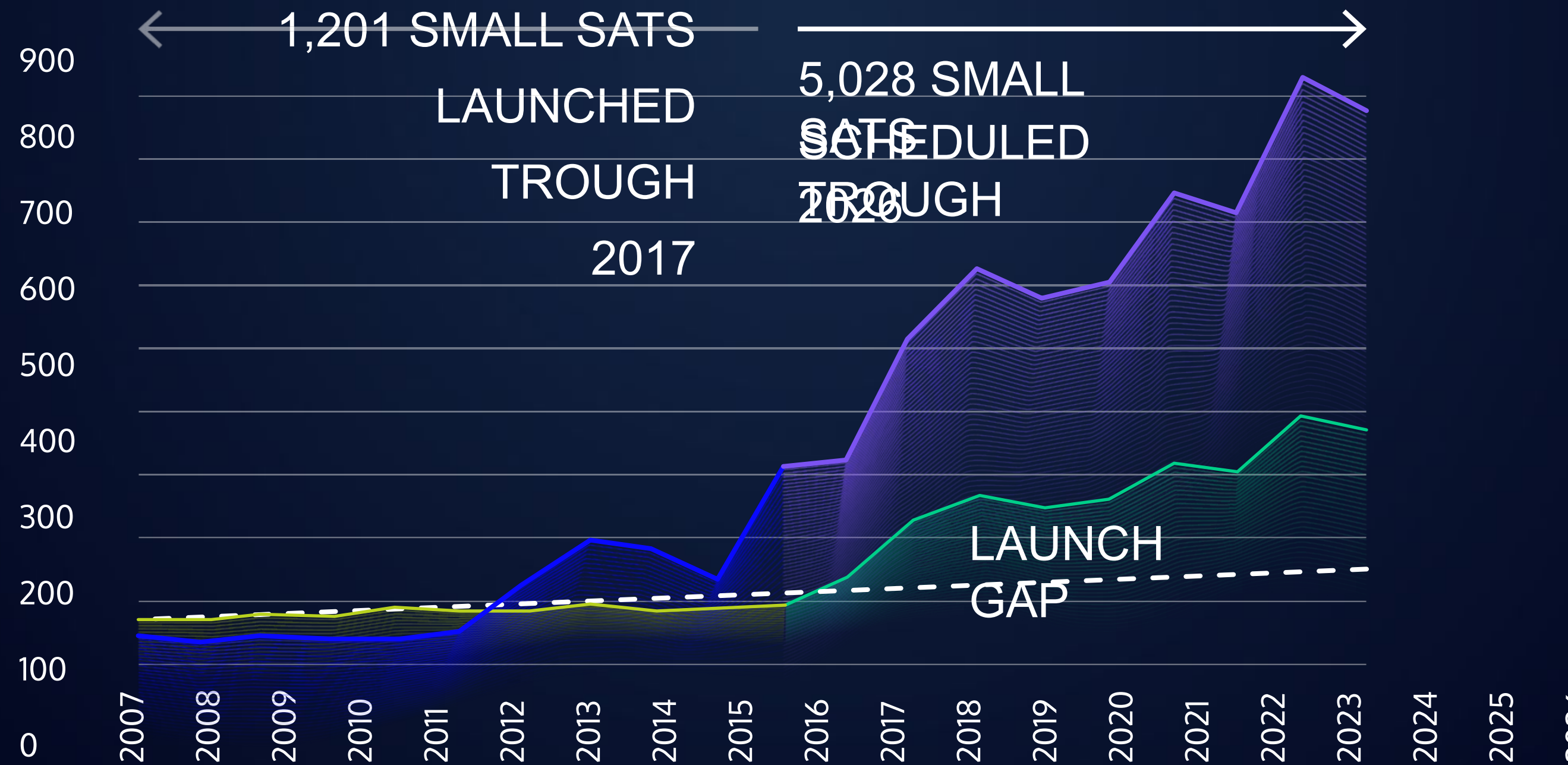
LAUNCH DEMAND

SMALL SATELLITE

DEMAND

- Small sats (Historical)
- Small sats (Forecasted)
- Total launches (Historical)
- Total launches needed (Forecast)
- Total launches (Trend)

Sources: Euroconsult (2017) / Firefly (2018)



TEA



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MBE**
Business strategist
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**TOM
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CEO and founder of
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MAIN CHARACTERISTICS OF

LV²⁴²⁰⁰



Propellants
 oxidizer:
 hydrogen peroxide
 fuel:
 kerosene

Name	1 stage	2 stage	3 stage	Type of main engine	1 stage	2 stage	3 stage
	(kg)	(kg)	(kg)		9x LPRE	1xLPRE	1xLPRE
Launch mass	55534	10284	778	Nominal thrust of main engine, N			
Payload fairing	190	190	190	- above sea level	70000		
Final mass	14217	2144	278	- in vacuum	80142	85500	3500
Mass of separating part	3933	1366	278	Nominal specific impulse, s			
Dry mass of the stage	3185	1110	225	- above sea level			
Mass of filled propellant	42065	8206	553	- in vacuum	250,4		
Usable propellant	41317	7950	500	Nozzle exit area, m ²	286,4	306	306,5
Ineffective propellant	748	256	53	Max g-load, units	0,099	0,471	0,124
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