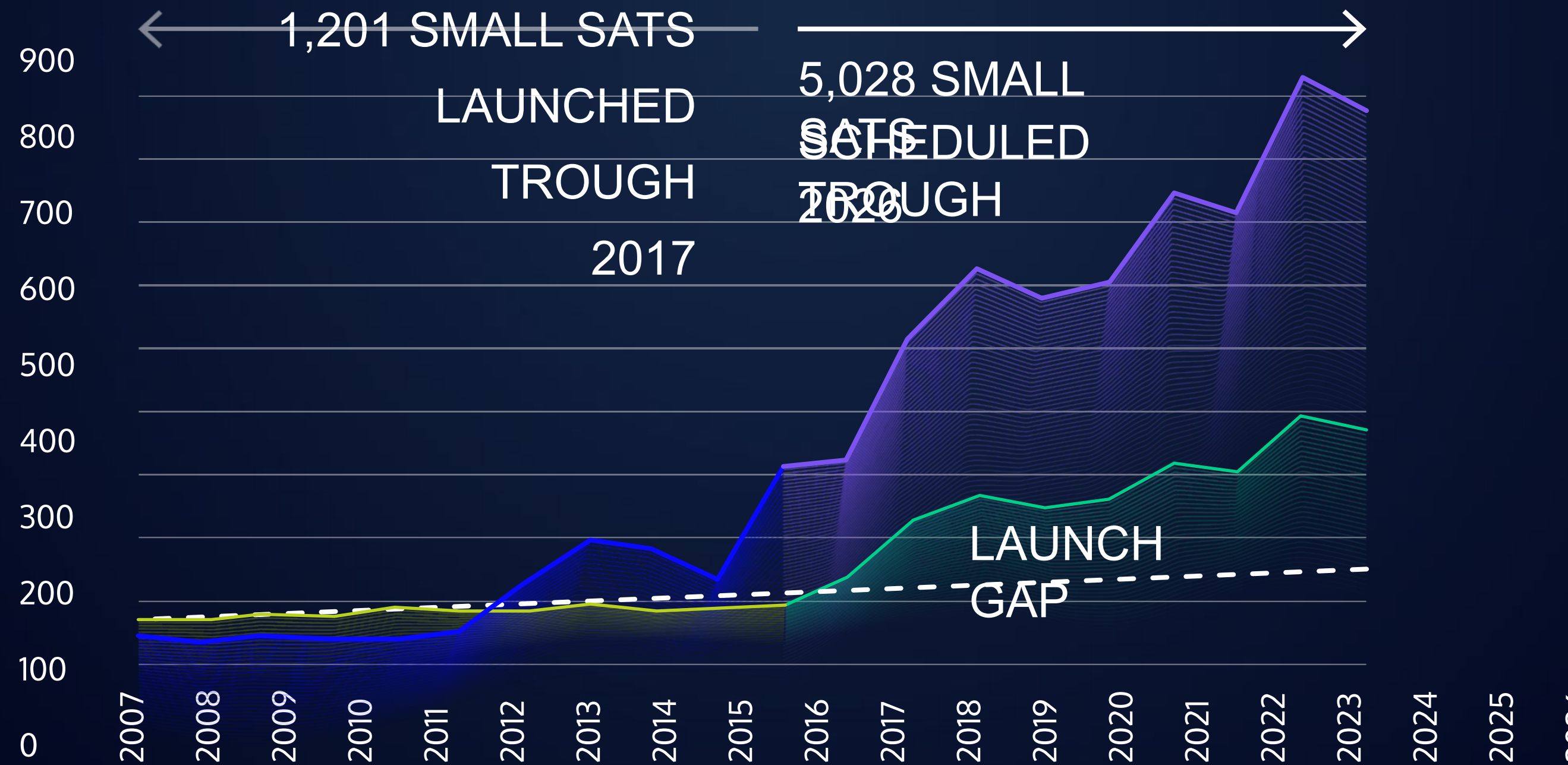


LAUNCH DEMAND SMALL SATELLITE

DEMAND

- Small sats (Historical)
- Small sats (Forecasted)
- Total launches (Historical)
- Total launches needed (Forecast)
- · - Total launches (Trend)

Sources: Euroconsult (2017) /
Firefly (2018)



TEA



**VLADIMIR
LEVYKIN**
CEO



**LAURA
EDISON**
Legal
Counsel



**DANIEL
SMITH**
Director of Business
Development



**PAVEL
BOTOULYK**
Co-Founder, Aurora UA



**RICHARD
OSBORN**
Programme Manager



**ROBIN
HAGE**
Engineer

ADVISOR BOARD



**GERRY
WEBB**
Co-founder and
Managing Director
of CST



**CRAIG CLARK,
MBE**
Business strategist
and founder of
Clyde Space



**TOM
MARKUSIC**
CEO and founder of
Firefly

MAIN CHARACTERISTICS OF

LV²⁴²⁰⁰



Propellants
oxidizer:
hydrogen peroxide
fuel:
kerosene

Name	1 stage	2 stage	3 stage	Type of main engine	1 stage	2 stage	3 stage
	(kg)	(kg)	(kg)		9x LPRE	1xLPRE	1xLPRE
Launch mass	55534	10284	778	Nominal thrust of main engine, N			
Payload fairing	190	190	190	- above sea level	70000		
Final mass	14217	2144	278	- in vacuum	80142	85500	3500
Mass of separating part	3933	1366	278	Nominal specific impulse, s			
Dry mass of the stage	3185	1110	225	- above sea level			
Mass of filled propellant	42065	8206	553	- in vacuum	250,4		
Usable propellant	41317	7950	500	Nozzle exit area, m ²	286,4	306	306,5
Ineffective propellant	748	256	53	Max g-load, units	0,099	0,471	0,124
					5	5	5